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Abstract : STATIC TESTS WERE CARRIED OUT ON AN 1/3 SCALE AIRCRAFT INTAKE MODEL PROVIDED BY ADA. THE TESTS (SUBSONIC) WERE CARRIED OUT USING AN EJECTOR DEVELOPED FOR THE PURPOSE. DUE TO LOSSES AT THE INTAKE ENTRANCE, IT WAS NOT POSSIBLE TO DRAW THE REQUIRED AIR FLOW THROUGH THE INTAKE. INTAKE WALL STATIC AND ENGINE FACE TOTAL PRESSURE DISTRIBUTIONS WERE DETERMINED FOR ENGINE FACE MACH NUMBERS VARYING FROM 0.17 TO 0.43. THE INTAKE PERFORMANCE, INCLUDING THE ENTRANCE EFFECTS FOR THE GEOMETRY GIVEN, HAS BEEN PRESENTED IN THIS REPORT, AS DETERMINED UNDER STATIC CONDITIONS.	